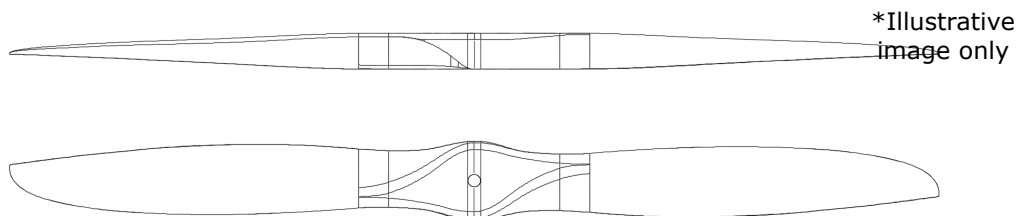


PROPELLER

23x8 2B GAS EVO

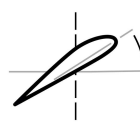
PN: 223086



138 g
Mass



23.0"
Diameter



8.0"
Pitch



Fixed wing

Engine/Motor type: Gas

Rotation direction: Counter-clockwise (Direction Guide)

Mass [g]: $138 \pm 4.0\%$

Moment of inertia [kgm^2]¹: $3.92\text{e-}03$

Core diameter [mm]: 40 (Drilling guide)

Limit RPM²: 7800

Working temperature [°C]: from -20°C to 60°C

Production method: Wet layup

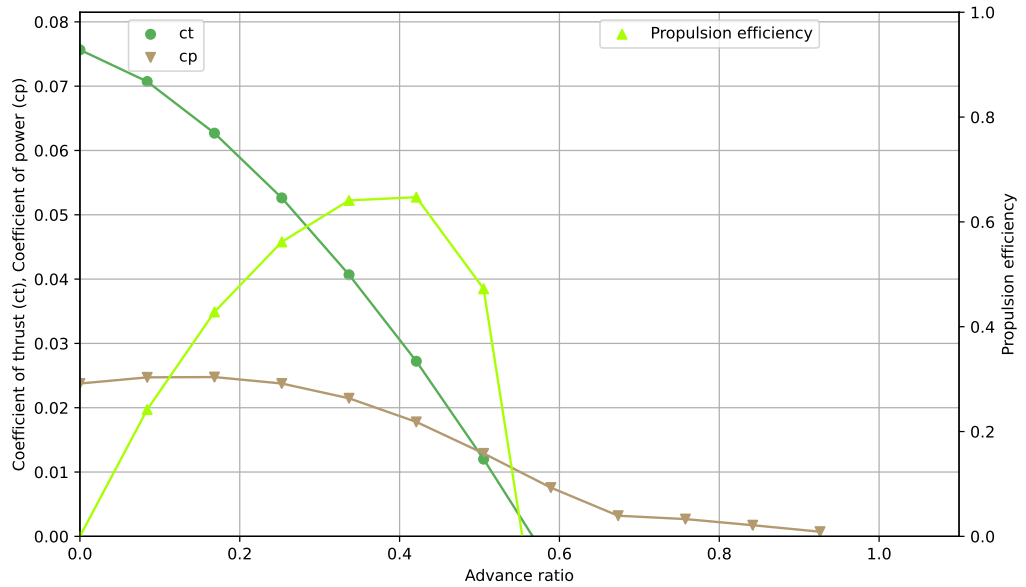
¹ Moment of inertia is only an estimation: $I = \frac{1}{24} \cdot \text{mass} \cdot \text{diameter}^2 \cdot n^{\circ} \text{ of blades}$

² RPM is limited by tip speed; forward speed reduces the limit.

For more information or custom propeller options, contact info@mejzlik.eu.
Operation manual: Propeller Maintenance and Repair Manual

Dynamic Performance Data

Simulated data - at 6100 RPM



v_inf	Ct	Cp	Propulsion efficiency	Advance ratio
0.0	0.0757	0.0238	0.0	0.0
5.0	0.0707	0.0247	0.2419	0.0842
10.0	0.0627	0.0248	0.4282	0.1684
15.0	0.0526	0.0238	0.5618	0.2526
20.0	0.0407	0.0215	0.641	0.3367
25.0	0.0272	0.0178	0.6471	0.4209
30.0	0.012	0.0129	0.4725	0.5051
35.0	-0.0045	0.0076	-0.3482	0.5893
40.0	-0.02	0.0032	-4.2202	0.6735
45.0	-0.0285	0.0027	-8.0639	0.7577
50.0	-0.0357	0.0017	-17.4578	0.8418
55.0	-0.0411	0.0007	-52.7796	0.926

$$C_T = \frac{T}{\rho R P S^2 D^4}$$

$$C_P = \frac{P_{mech}}{\rho R P S^3 D^5}$$

$$\eta = \frac{C_T \cdot J}{C_P}$$

$$J = \frac{v_{inf}}{n \cdot D}$$