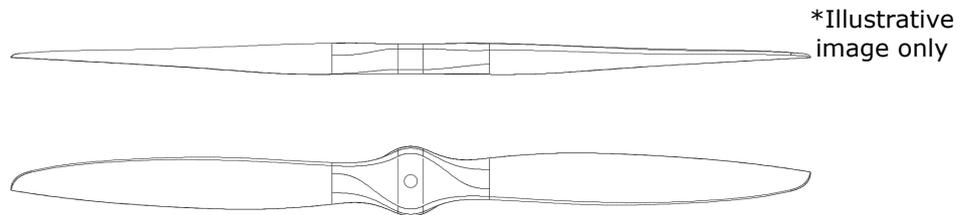


# PROPELLER

## 26.5x16.2 2B GAS

PN: 226163



192 g  
Mass



26.5"  
Diameter



16.2"  
Pitch



Fixed wing

Engine/Motor type: Gas

Rotation direction: Clockwise (Direction Guide)

Mass [g]:  $192 \pm 10.0\%$

Moment of inertia [ $\text{kgm}^2$ ]<sup>1</sup>:  $7.25\text{e-}03$

Core diameter [mm]: 46 (Drilling guide)

Limit RPM<sup>2</sup>: 6800

Working temperature [°C]: from -20°C to 60°C

Production method: Wet layup

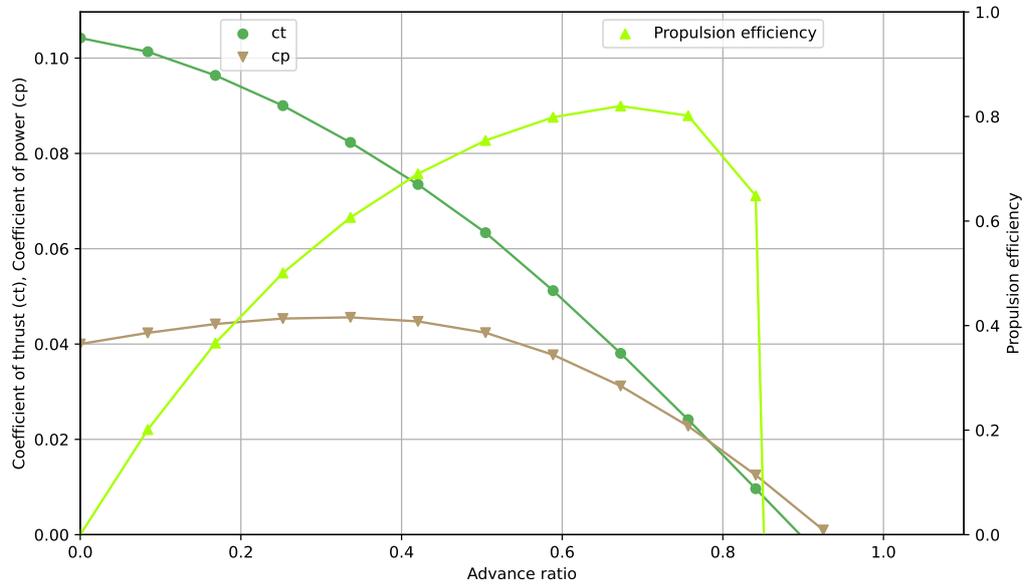
<sup>1</sup> Moment of inertia is only an estimation:  $I = \frac{1}{24} \cdot mass \cdot diameter^2 \cdot n^{\circ}ofblades$

<sup>2</sup> RPM is limited by tip speed; forward speed reduces the limit.

For more information or custom propeller options, contact [info@mejzlik.eu](mailto:info@mejzlik.eu).  
Operation manual: Propeller Maintenance and Repair Manual

# Dynamic Performance Data

Simulated data - at 5300 RPM



v_inf	Ct	Cp	Propulsion efficiency	Advance ratio
0.0	0.1042	0.04	0.0	0.0
5.0	0.1013	0.0423	0.2012	0.0841
10.0	0.0964	0.0442	0.3666	0.1682
15.0	0.09	0.0454	0.5007	0.2523
20.0	0.0823	0.0456	0.607	0.3364
25.0	0.0735	0.0447	0.6904	0.4205
30.0	0.0634	0.0424	0.7542	0.5046
35.0	0.0512	0.0377	0.7987	0.5887
40.0	0.0381	0.0312	0.8201	0.6728
45.0	0.0241	0.0228	0.8016	0.7568
50.0	0.0096	0.0125	0.6484	0.8409
55.0	-0.005	0.001	-4.7379	0.925

$$C_T = \frac{T}{\rho RPS^2 D^4}$$

$$C_P = \frac{P_{mech}}{\rho RPS^3 D^5}$$

$$\eta = \frac{C_T \cdot J}{C_P}$$

$$J = \frac{v_{inf}}{n \cdot D}$$